

## ABSTRACT OF THE DISCLOSURE

A thermal barrier coating enabling to prevent peeling of layer in a high temperature use and still having a high thermal barrier effect, a turbine part coated by this thermal barrier coating and a gas turbine comprising this turbine part are provided. The thermal barrier coating comprises a base material 21 of high temperature heat resistant alloy and a ceramics layer 23 formed on the base material 21. The ceramics layer 23 comprises  $ZrO_2$  added with  $Yb_2O_3$  as stabilizer and is laminated on the base material via a bond coat layer 22 laminated as a metallic bond layer. A plurality of cracks 23A are preferably introduced in the ceramics layer 23. The turbine part is constructed having its surface coated with the above thermal barrier coating.